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(Meredith Hoppe)

## GESAMMELTE SCHRIFTEN IN FUNF BANDEN (4)



RareBooksClub. Paperback. Book Condition: New. This item is printed on demand. Paperback. 18 pages. Original publisher: Hampton, Va.: National Aeronautics and Space Administration, Langley Research Center, 1990 OCLC Number: (OCoLC)25698866 Subject: Airplanes -- Control systems. Excerpt: . . . CONCLUDING REMARKS This paper has described an aeroservothermoelastic analysis and design methodology for the implementation of active control law function. , , to hypersonic aircraft. With this methodology, the aerodynamic heat loads for hypersonic flight are determined and applied to finite element models of the aircraft structure to determine stiffness changes due to thermal stresses and material property changes. Using the hot structure vibration modes, an aeroelastic analysis is performed with Piston Theory unsteady aerodynamic forces to determine the thermal effects on flutter dynamic pressure. Once the flutter characteristics of the hot structure are known, active control functions such as flutter suppression, gust load alleviation, and ride quality improvement systems can be designed. These systems can overcome the adverse dynamic response characteristics brought on by the therrnoelastic destiffening of the aircraft structure. CONCLUDING REMARKS Aerothermoelastic analysis capability available-APAS HABP used for stability derivatives, aerodynamic center locations, and aerodynamic heat loads-Heat loads applied to finite element structural model for hoVcold vibration modes-Nonrigid chord Piston Theory unsteady aerodynamics coupled with quasi-steady aerodynamic approximation method Active controls can compensate for thermoelastic effects-Thermoelastic effects result in lower flutter boundaries-Twenty-third order LQG FSS control law more than recovered cold structure flutter boundary using normal acceleration feedbacks Pilot station ride quality in turbulence improved using full-state feedback gain to place closed-loop eigenvalues-Effects of control system nonlinearities yet to be determined by simulation 14 This item s



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